



**TYPE CERTIFICATE**

**EASA.A.576**

This Type Certificate is issued by EASA, acting in accordance with Regulation (EC) No. 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation and in accordance with Commission Regulation (EC) No. 748/2012 to

**COSTRUZIONI AERONAUTICHE TECNAM  
S.R.L.**

VIA TASSO 478  
80127 NAPOLI NA  
ITALY

and certifies that the product type design listed below complies with the applicable Type Certification Basis and environmental protection requirements when operated within the conditions and limitations specified on the associated:

Type Certificate Data Sheet Number: EASA.A.576

**Type Design - Model**

P2010

**Date of Issue**

26 September 2014

**For the European Aviation Safety Agency,**

**Date of issue: 26 September 2014**

  
**Yves MORIER**  
**Head of General Aviation and**  
**Remotely Piloted Aircraft Systems (RPAS)**



## ***European Aviation Safety Agency***

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**EASA**

**TYPE-CERTIFICATE  
DATA SHEET**

**EASA.A.576**

**P2010**

**Costruzioni Aeronautiche TECNAM S.r.l.**

Via Tasso, 478  
80127 Napoli  
ITALIA

Issue 01: 26 Sept 2014

Issue 02: 05 May 2015

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## **SECTION A: P2010**

### **A.I. General**

- |   |  |
|---|--|
| 1. Data Sheet No.:                                      | EASA.A.576   |
| 2. a) Type:   | P2010  |
| 3. Airworthiness Category:                              | CS-23 Normal category  |
| 4. Type Certificate Holder:                             | Costruzioni Aeronautiche Tecnam S.R.L.<br>Via Tasso, 478<br>80127 Napoli<br>ITALIA |
| 5. Manufacturer:  | Costruzioni Aeronautiche Tecnam S.R.L.<br>Via Tasso, 478<br>80127 Napoli<br>ITALIA |
| 6. Certification Application Date:                      | 15 September 2010  |
| 7. (Reserved) National Certifying Authority             | N/A  |
| 8. (Reserved) National Authority Type Certificate Date: | N/A  |

### **A.II. EASA Certification Basis**

- |  |  |
|--|--|
| 1. Reference Date for determining the applicable requirements: | 15 September 2010  |
| 2. Airworthiness Requirements:                                 | EASA CS-23 amdt.2 dated 28 September 2010  |
| 3. Special Conditions:   | CRI B-52 (SC-B23.div-01 Human Factors – Integrated Avionic System);<br>CRI F-101 (SC-F23-1309-02 Protection from the Effect of HIRF);<br>CRI F-54 (SC-F23-1309-03 Protection from the Effects of Lightning Strike, Indirect Effects);<br>CRI F-58 (SC-F23.1353-02 Lithium Battery Installations)<br>CRI F-102 (ESP and USP) (see note 3) |
| 3. Exemptions:   | None   |
| 4. Deviations:   | None   |
| 5. Equivalent Safety Findings:                                 | None   |
| 6. Requirements elected to comply:                             | None   |

7. Environmental Standards: CS-36 amdt. 2 dated 31 August 2009, subpart C with reference to ICAO Annex 16, Volume 1, Chapter 10, amdt. 9 dated 30 July 2009.
8. (Reserved) Additional National Requirements: N/A
9. (Reserved) N/A

### **A.III. Technical Characteristics and Operational Limitations**

1. Type Design Definition: Document no. 2010/010 "Type Design Definition"
2. Description:
- 2.1 Basic: Single-engine, fixed pitch propeller, four seats, high wing aeroplane equipped with fixed tricycle landing gear, featuring composite, aluminium and steel construction.
- 2.2 Optional (see note 1) Single-engine, variable pitch propeller, four seats, high wing aeroplane equipped with fixed tricycle landing gear, featuring composite, aluminium and steel construction.
3. Equipment: Equipment list, AFM, doc. No. 2010/100, Section 6
4. Dimensions:
- |           |  |
|-----------|--|
| Span      | 10.30 m (33.79 ft)                           |
| Length    | 7.97 m (26.15 ft)                            |
| Height    | 2.64 m (8.66 ft)                             |
| Wing Area | 13.9 m <sup>2</sup> (149.6 ft <sup>2</sup> ) |
5. Engine:
- 5.1.1 Model: No.1 Lycoming Engines: IO-360-M1A
- 5.1.2 Type Certificate: EASA Type Certificate No. EASA.IM.E.032
- 5.1.3 Limitations
- 5.1.3.1 Basic: Take-Off Power 134 kW (180HP) at 2700 RPM  
Max continuous power 134 kW (180HP) at 2700 RPM  
Other engine's limitations are listed in doc. No. 2010/100 "P2010 Aircraft Flight Manual", Section 2
- 5.1.3.2 Optional (see note 1) Take-Off Power 134 kW (180HP) at 2700 RPM  
Max continuous power 129 kW (173HP) at 2600 RPM  
Other engine's limitations are listed in doc. No. 2010/100 "P2010 Aircraft Flight Manual", Section 2
6. Load factors:
- |          | Flap UP | Flap DOWN |
|----------|---------|-----------|
| Positive | +3.8 g  | +2.0 g    |
| Negative | -1.52 g | 0.0 g     |

7. Propeller:

7.1 Basic:

- 7.1.1 Model: MT Propeller: MT 188 R 145-4G
- 7.1.2 Type Certificate: EASA Type Certificate No. EASA.P.006
- 7.1.3 Number of blades: 2
- 7.1.4 Diameter: 1.880 m (74 in) – No reduction is permitted
- 7.1.5 Sense of Rotation: Clockwise (pilot’s view)

7.2 Optional:

(see note 1)

- 7.2.1 Model: MT Propeller: MTV-15-B/193-52
- 7.2.2 Type Certificate: EASA Type Certificate No. EASA.P.098
- 7.2.3 Number of blades: 2
- 7.2.4 Diameter: 1.930 m (76 in) – No reduction is permitted
- 7.1.5 Sense of Rotation: Clockwise (pilot’s view)

- 8.1 Fuel: AVGAS Grade 91/96 or 100 LL (ASTM D910)  
MOGAS EN 228:2008 (E) (see note 2)  
Refer to doc. No. 2010/100 “P2010 Aircraft flight Manual” for further details.

8.2 Oil:

Average Ambient Temperature	MIL-L-6082B or SAEJ1966 Spec. Mineral Grades	MIL-L-22851 or SAEJ1899 Spec. Ashless Dispersant Grades
All Temperatures	----	SAE15W50 or SAE20W-50
Above 80°F	SAE60	SAE60
Above 60°F	SAE50	SAE40 or SAE50
30°F to 90°F	SAE40	SAE40
0°F to 70°F	SAE30	SAE40, SAE30, SAE20W40
Below 10°F	SAE20	SAE30 or SAE20W30

Refer to Lycoming (L)LIO-360-M1A “Operation and Installation Manual” for list of alternative recommended commercial brands and types.

9. Fluid capacities:

9.1 Fuel:	2 Tanks:	120 litres each (31.7 US gallons)
	Total:	240 litres (63.4 US gallons)
	Usable:	231 litres (61 US gallons)
9.2 Oil:	Total:	7.57 litres (8 US qts)
	Minimum:	3.78 litres (4 US qts)

10. Air Speeds:

Never exceed speed $V_{NE}$	164 KCAS
Maximum Structural Cruising Speed $V_{NO}$	130 KCAS
Design Manoeuvring speed $V_A$	119 KCAS
Operating Manoeuvring speed $V_O$	119 KCAS
Maximum flaps extended speed $V_{FE}$	92 KCAS

11. Maximum Operating Altitude:

12000 ft

12. Allweather Operations Capability:

Day/Night-VFR, IFR ;  
Refer to KOEL contained in the AFM, doc. No. 2010/100, Section 2.  
Flight into expected or actual icing conditions is prohibited

13. Maximum Weights:

Max Take-Off:	1160 kg (2557 lb)
Max Landing:	1160 kg (2557 lb)

14. Centre of Gravity Range:

Forward Limit: 0.262 m (19% MAC) behind datum  
Aft Limit: 0.440 m (32% MAC) behind datum  
Mean Aerodynamic Chord is 1.378 m (54.2 in)

15. Datum:

Vertical plane tangent to wing leading edge

16. Control surface deflections:

Stabilator:  $17^{\circ}\pm 2^{\circ}$  to pitch up /  $6^{\circ}\pm 2^{\circ}$  to pitch down  
Stabilator Trim Tab:  $15^{\circ}\pm 1^{\circ}$  downward /  $3^{\circ}\pm 1^{\circ}$  upward  
Aileron:  $19^{\circ}\pm 2^{\circ}$  upward /  $14^{\circ}\pm 2^{\circ}$  downward  
Rudder:  $25^{\circ}\pm 2^{\circ}$  left /  $25^{\circ}\pm 2^{\circ}$  right  
Rudder Trim Tab:  $20^{\circ}\pm 2^{\circ}$  left /  $20^{\circ}\pm 2^{\circ}$  right  
Flaps:  $0^{\circ}$  Fully Retracted /  $40^{\circ}\pm 1^{\circ}$  Fully Extended

17. Levelling Means:

seat track supporting beams (see procedure in doc. No. 2010/100 "P2010 Aircraft Flight Manual", Section 6)

18. Minimum Flight Crew: 1
19. Maximum Passenger Seating Capacity: 3
20. Baggage/Cargo Compartments: Max Allowable Load: 40 kg (88 lb)  
Location:1.56 m (61.41 in) from datum
21. Wheels and Tyres: Nose Wheel Tyre Size: 5.00-5, Type III  
Main Wheel Tyre Size 6.00-6, Type III  
For approved Types and rating see AMM, doc No. 2010/101
22. Serial Numbers Eligible: 002 to subsequent



#### **A.IV. Operating and Service Instructions**

1. Flight Manual: Doc. No. 2010/100 "P2010 Aircraft Flight Manual"  
Last issue.
2. Technical Manual: Doc. No. 2010/101 "P2010 Aircraft Maintenance  
Manual" Last issue;  
Airworthiness Limitations are reported in ATA  
chapter 4.
3. Spare Parts Catalogue: Doc. No. 2010/102 "P2010 Illustrated Parts  
Catalogue" Last issue.
4. Instruments and aggregates: Doc. No. 2010/101 "P2010 Aircraft Maintenance  
Manual" Last issue.

**A.V. Notes:**

- 1) When MOD 2010/002 (EASA approval 10052750) is installed
- 2) When MOD 2010/032 (EASA approval 10055692) is installed
- 3) When MOD 2010/001 (EASA approval 10055187) is installed

## **ADMINISTRATIVE SECTION**

### I. Acronyms

AFM – Aircraft Flight Manual  
 AMM – Aircraft Maintenance Manual  
 ASTM – American Society for Testing and Materials  
 CRI – Certification Review Item  
 CS – Certification Specification  
 EASA – European Aviation Safety Agency  
 ICAO – International Civil Aviation Organization  
 IPC – Illustrated Part Catalogue  
 KCAS – Knots Calibrated Air Speed  
 KOEL – Kind of Operations Equipment List  
 MAC – Mean Aerodynamic Chord  
 MLW – Maximum Landing Weight  
 MTOW – Maximum Take-Off Weight  
 MZFW – Maximum Zero Fuel Weight  
 TC – Type Certificate  
 TCDS – Type Certificate Data Sheet  
 VFR – Visual Flight Rules  
 IFR – Instrumental Flight Rules

### II. Type Certificate Holder Record

<b>TC Holder</b>	<b>Period</b>
Costruzioni Aeronautiche TECNAM S.r.l. Via Tasso, 478 80127 Napoli ITALIA	Effective

### III. Change Record

<b>Issue</b>	<b>Date</b>	<b>Changes</b>	<b>TC Issue No. &amp; Date</b>
Issue 01	26 Sept 2014	Initial Issue	26 Sept 2014
Issue 02	05 May 2015	MT Variable Pitch Propeller Added	
Issue 03	15 Dec 2015	Update to include changes: MOD2010/001 "GFC 700 autopilot" (EASA approval 10055187), MOD2010/003 "Alternative avionics configuration" (EASA approval 10053996), MOD2010/032 Automobile fuel (EASA approval 10055692)	